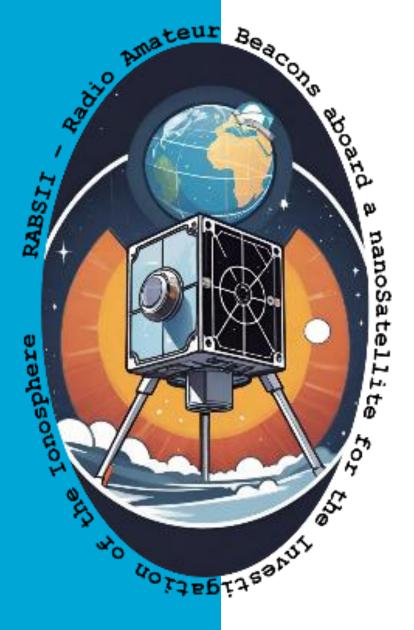


RABSII: Radio Amateur Bakens aan boord van een nanoSatelliet voor onderzoek van de Ionosfeer

JURGEN VANHAMEL — ON5ADL









Jurgen Vanhamel – ON5ADL

j.a.m.vanhamel@tudelft.nl jurgen.vanhamel@kuleuven.be



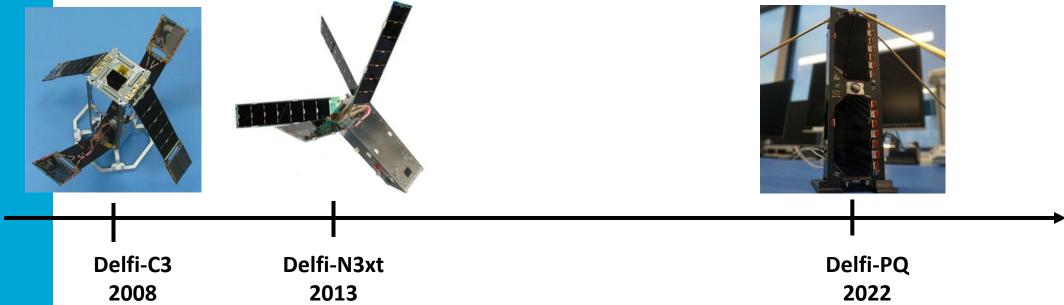






The Delfi Space Program

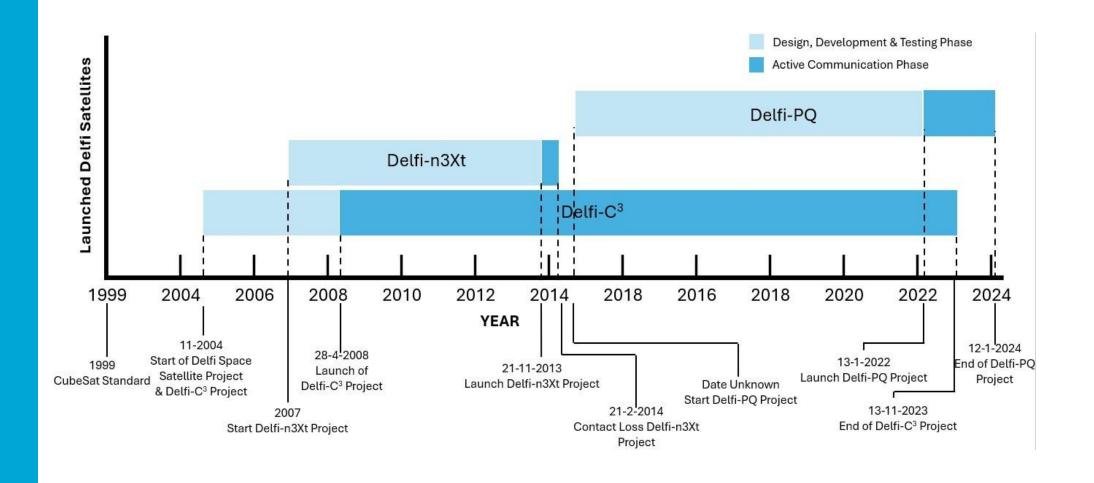
The Delfi Space program realizes education and research by the end-to-end engineering of space missions of high relevance and impact using very small satellites.







The Delfi Space Program

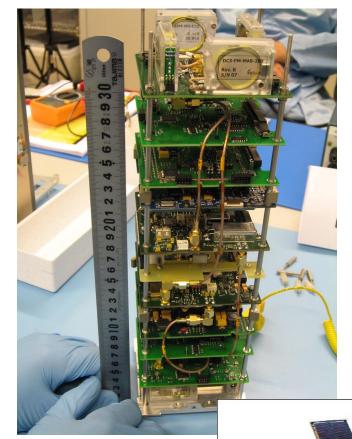






Delfi-C3

- 3U CubeSat
- Power subsystem w/o batteries
- AOCS using magnetic-damping
- Passive thermal subsystem
- Decentralized CDHS
- Decentralized Electrical Power System
- Uplink UHF @ 435 MHz
- Downlink VHF @ 145 MHz, 1200 bps







Delfi-n3Xt

- 3U CubeSat
- Fully redundant OBC
- Single Point of Failure free power system
- 30 Wh battery system
- 3-axis attitude control
- Failure tolerant on-board data storage
- Uplink UHF @ 435 MHz
- Downlink VHF @ 145 MHz, 1200 bps
- Downlink S-band @ 2.4 GHz > 10 250 kbps

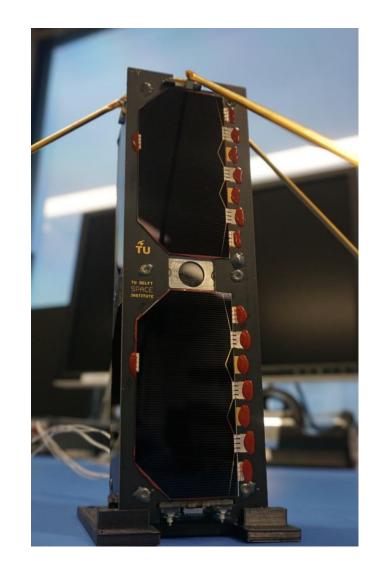






Delfi-PQ

- 3P PocketQube (178x50x50 mm)
- Single string modular bus design
- Passive temperature control
- Similar Performances to Delfi-C3/N3xt
- Magnetic attitude control
- Full sw update capabilities (lab in space)
- Uplink UHF @ 145 MHz
- Downlink VHF @ 435 MHz, 9600 bps

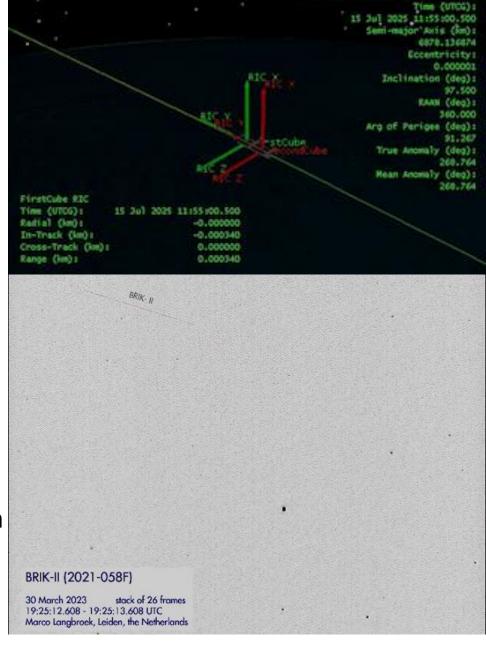






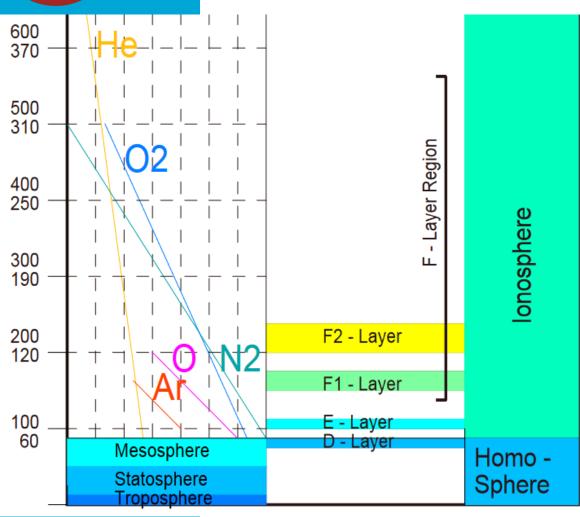
New mission!

- Satellite formation
 - Two (or more?) satellites
- Educational purpose
 - Lab in space
 - Direct involvement in educational activities
 - STEM activities (local and worldwide)
- Scientific purpose
 - Precise orbit determination
 - GNSS + radiometric + laser tracking
 - Space Situational Awareness demonstration
 - Multi (ground) sensor detection
 - Sporadic E detection









Ionosphere → Composition

He: Helium

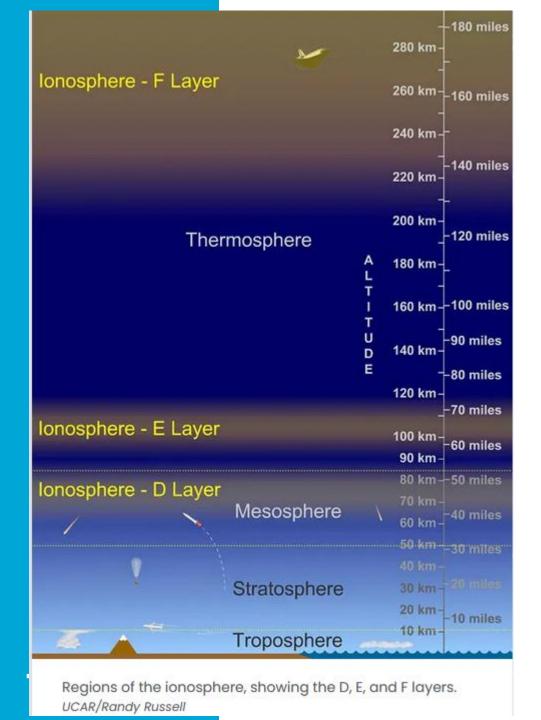
O: oxygen

O₂: chemically bound oxygen (oxygen molecule)

Ar: Argon

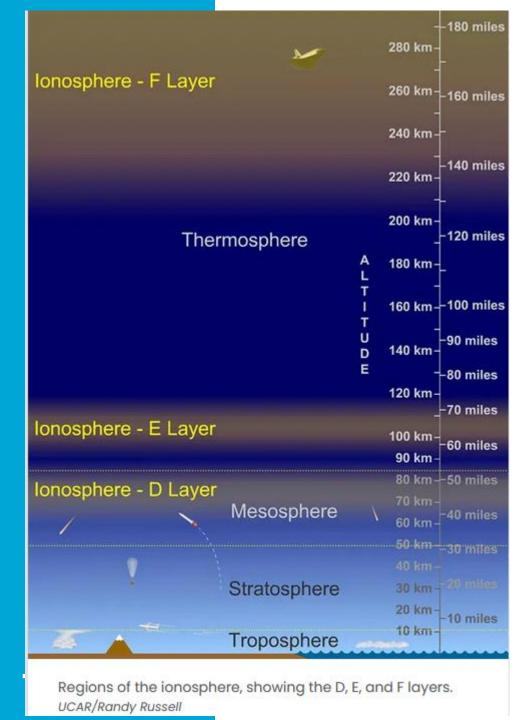
N₂: Nitrogen







Ionosphere → setup

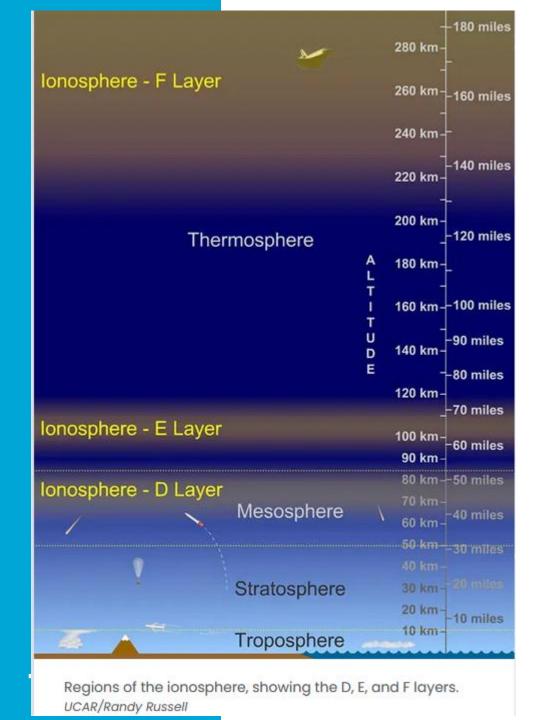




Ionosphere → setup

Ionisation process:

- **Sun's ultraviolet light** → ionizes atoms and molecules in the Earth's upper atmosphere (50-1000 km)
- X-ray and gamma-rays → produced when solar events (solar flares) occur
- → increase the density of the ionosphere on the dayside
- **Protons and electrons** → produced by solar events
- → precipitate into the ionosphere in the polar regions
- → produce increases in the ionosphere density at low altitudes
- Particles from the solar wind and cosmic radiation

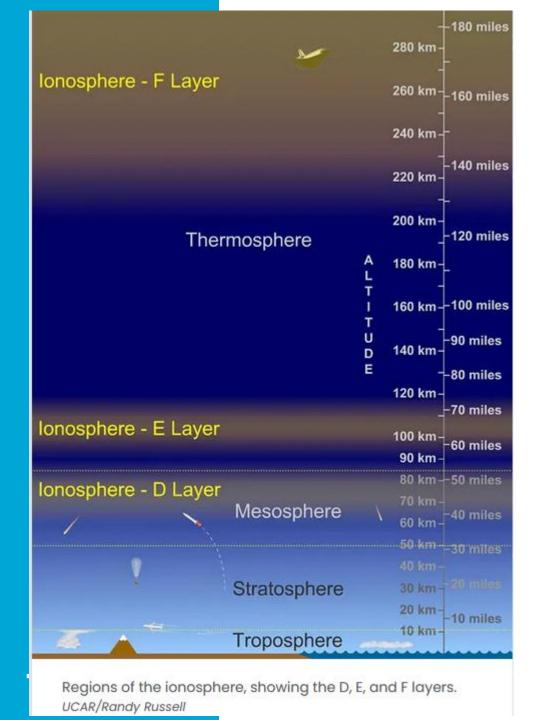


RF Seminar

lonosphere → setup

Ionisation process:

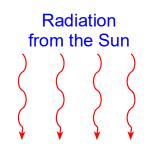
REGION	PRIMARY IONISING RADIATION FORMS
D	Lyman alpha, Hard X-Rays
Е	Soft X-Rays and some Extreme Ultra-Violet
F1	Extreme Ultra-violet, and some Ultra-Violet
F2	Ultra-Violet

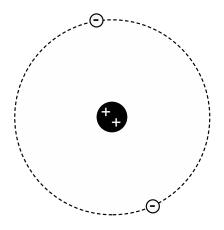


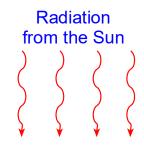


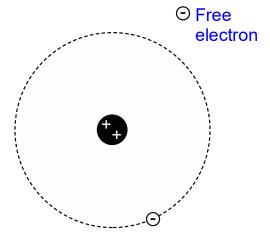
lonosphere → setup

Ionisation process:



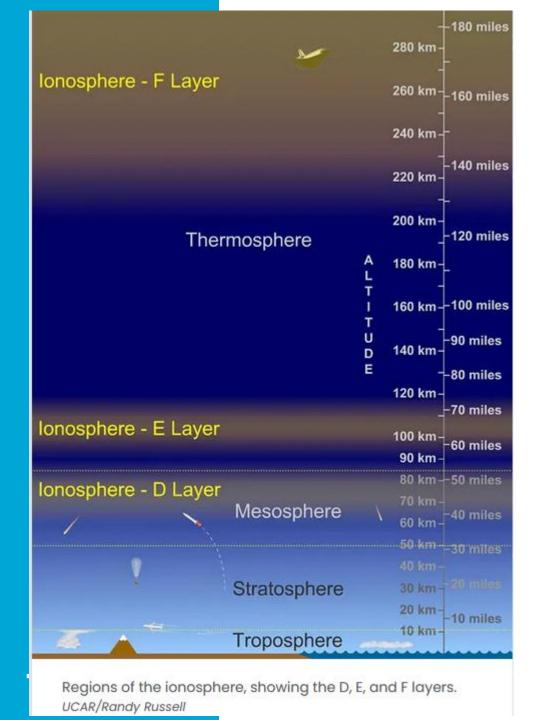






Molecule with all electrons

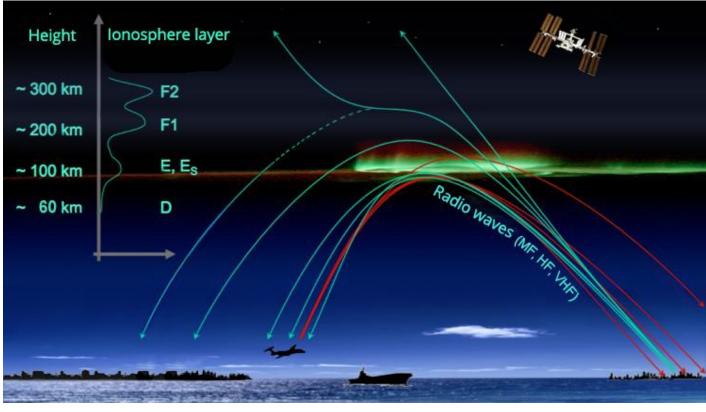
Positive ion with one electron short

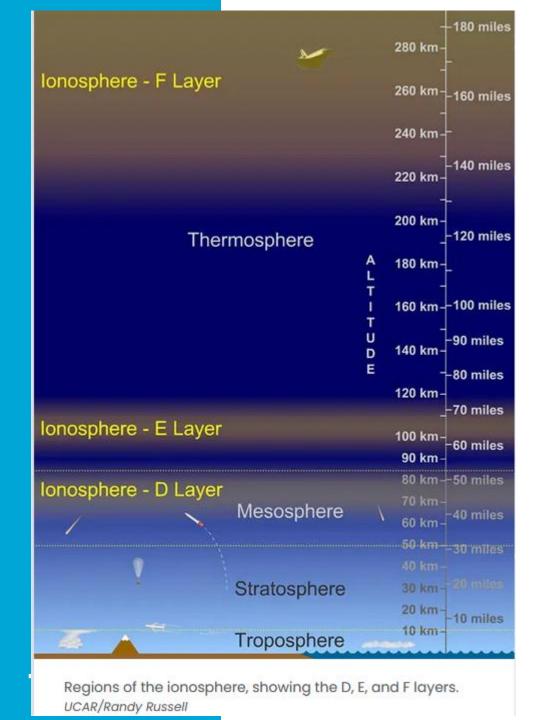


RF Seminar

lonosphere → setup

Radio wave reflection







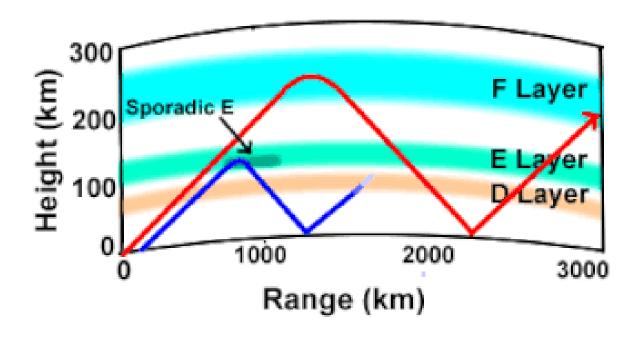
lonosphere → setup



Interest in the E layer region → sporadic E



Sporadic E?

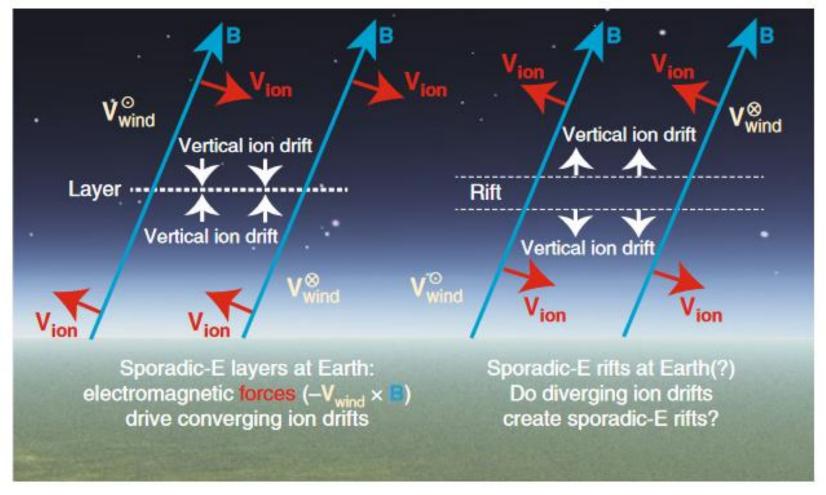


- Variability in size
- Variability in location
- Variability in timing
- Clear model is still lacking





Sporadic E?



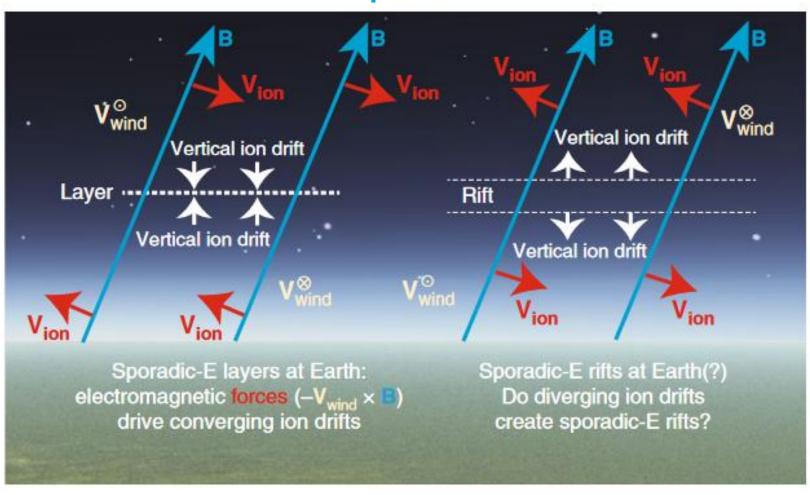


Collinson, G.A., McFadden, J., Grebowsky, J. et al. Constantly forming sporadic E-like layers and rifts in the Martian ionosphere and their implications for Earth. Nat Astron 4, 486–491 (2020)



Sporadic E?

Inclarities remain

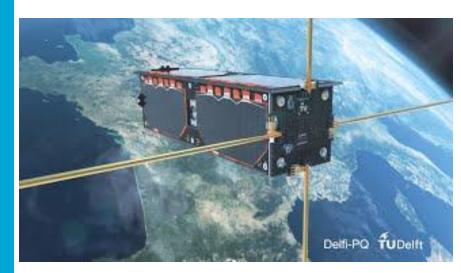


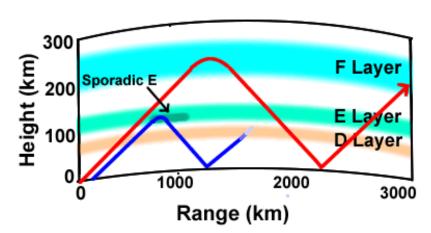


Collinson, G.A., McFadden, J., Grebowsky, J. et al. Constantly forming sporadic E-like layers and rifts in the Martian ionosphere and their implications for Earth. Nat Astron 4, 486–491 (2020)



What is a beacon system?



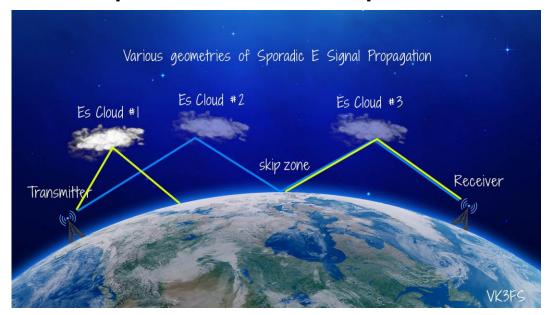




Scientific purpose of RABSII ??



Detect sporadic E in the ionosphere



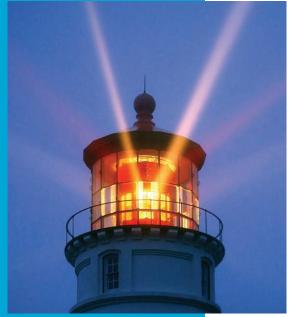


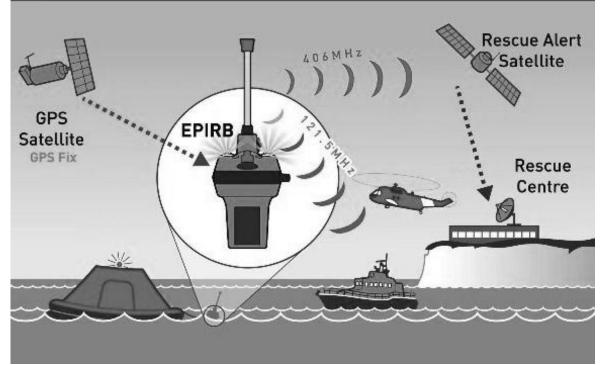


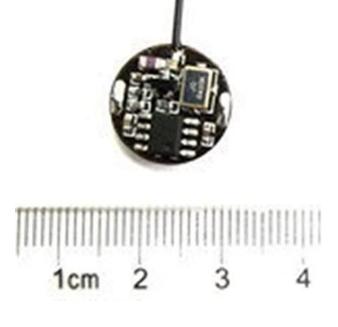
What is a beacon system?



Send info using a transmitter and antenna system











Implementing a CW and FT4 beacon aboard a small satellite



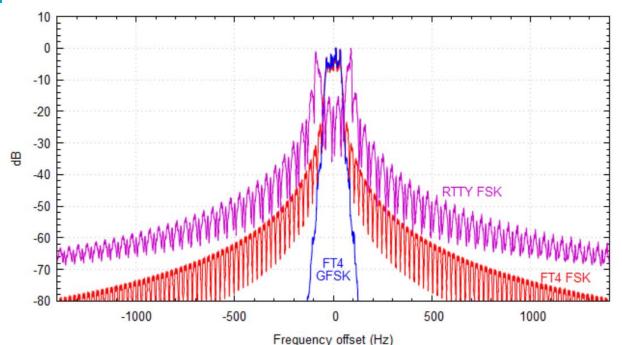


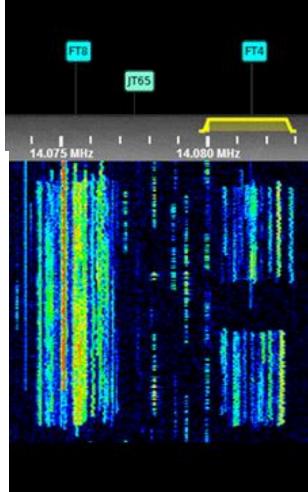
Implementing a CW and FT4 beacon aboard a small satellite



Working principle of FT4:

- 4-tone GFSK modulation;







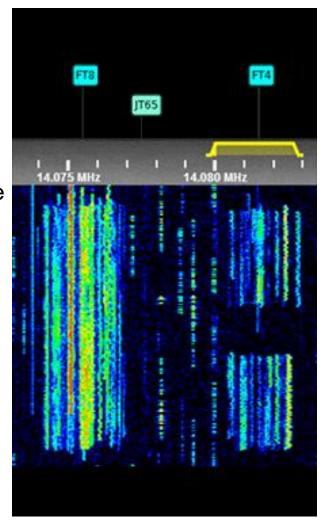


Implementing a CW and FT4 beacon aboard a small satellite



Working principle of FT4:

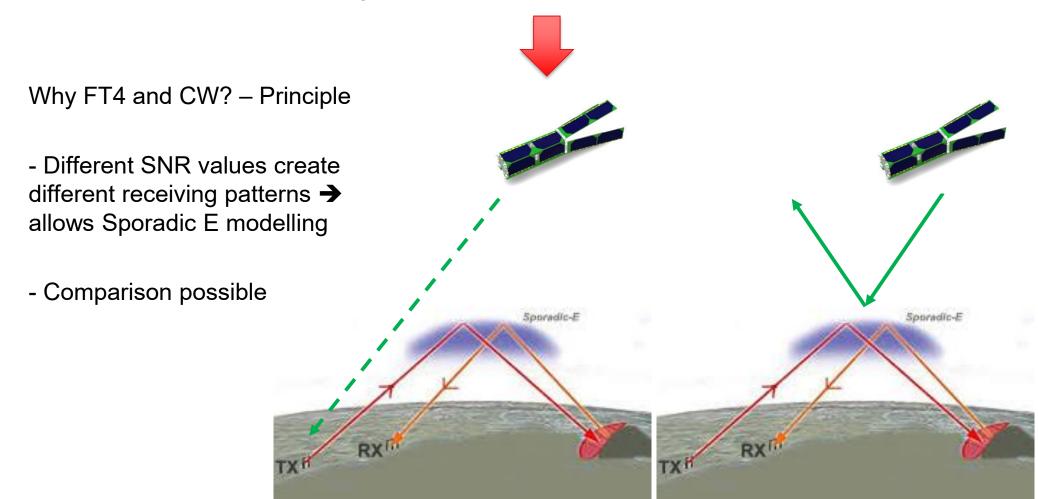
- 4-tone GFSK modulation;
- 6 seconds sequence: 4.48 s transmit, 1.52 s decode
- Tone separation 20,833 Hz;
- Bandwidth of 90 Hz;
- 50% decoding probability is S/N = -16.4 dB







Implementing a CW and FT4 beacon aboard a small satellite







Implementing a CW and FT4 beacon aboard a small satellite

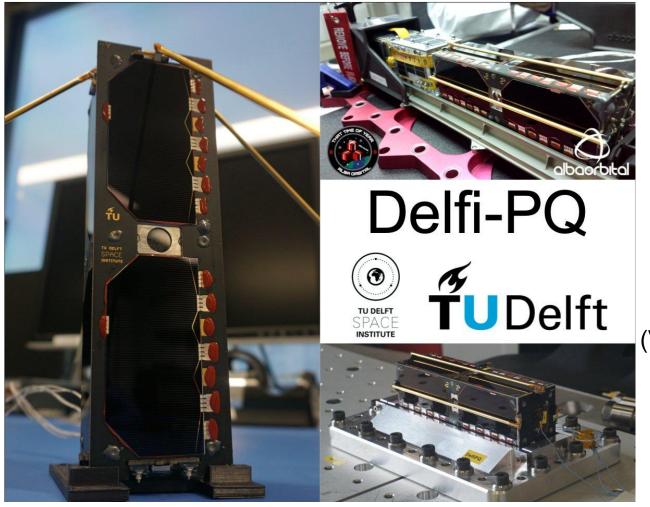


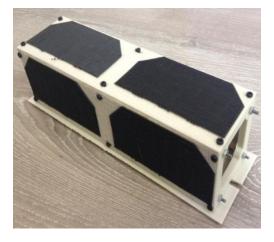
Implementation aboard the satellite:

160m	
80m	3.575
60m	
40m	7.0475
30m	10.140
20m	14.080
17m	18.104
15m	21.140
12m	24.919
10m	28.180
6m	50.318









Launch of Delfi-PQ



(Very) small satellites in space are hot!

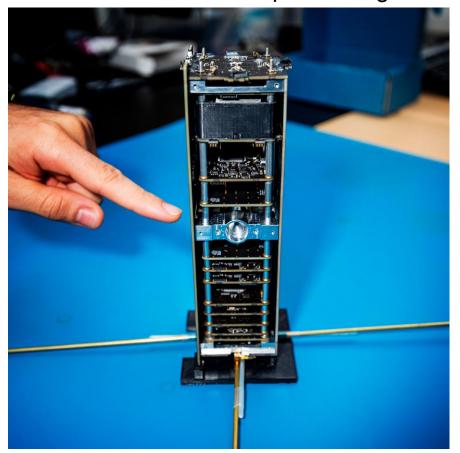


Many things still to discover



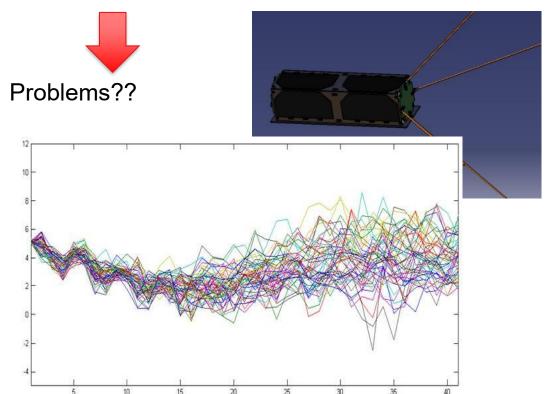


Implementing a CW and FT4 beacon aboard this small satellite





A transmitter & antenna are needed aboard the S/C



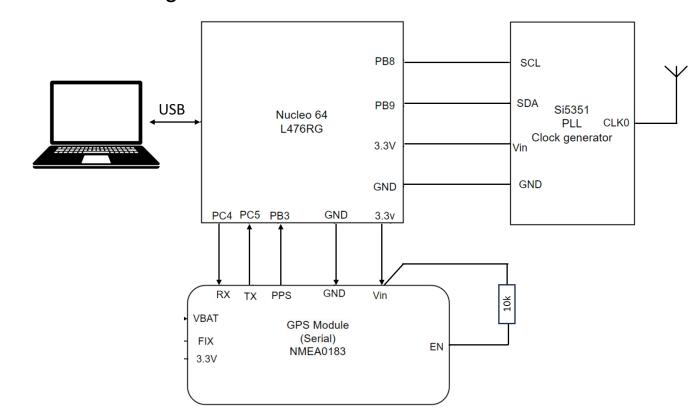




Implementing a CW and FT4 beacon aboard this small satellite



Design based on:





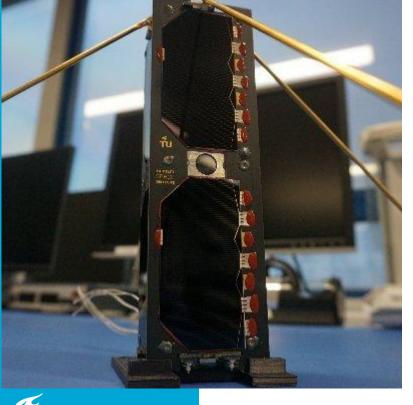




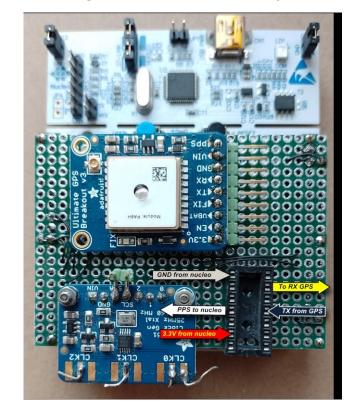
Implementing a CW and FT4 beacon aboard this small satellite



In-house designed instrument by RST-UBA / TU Delft









Implementing a CW and FT4 beacon aboard this small satellite



Testing:

Doppler shift LEO satellite:

$$f = f_0 \left(1 + \frac{v}{c} \right)$$

v = speed of the approaching satellite 7823 m/s c = velocity of the speed of light 3 . 10^8 m/s f_0 = 28 . 10^6 Hz





Implementing a CW and FT4 beacon aboard this small satellite



Testing:

Doppler shift LEO satellite:

$$f = f_0 \left(1 + \frac{v}{c} \right)$$

v = speed of the approaching satellite 7823 m/s c = velocity of the speed of light 3 . 10^8 m/s f_0 = 28 . 10^6 Hz

→ frequency on ground f = 28,000,730.17 Hz





Implementing a CW and FT4 beacon aboard this small satellite



Testing:

Doppler shift LEO satellite:

$$f = f_0 \left(1 + \frac{v}{c} \right)$$

v = speed of the approaching satellite 7823 m/s c = velocity of the speed of light 3 . 10⁸ m/s $f_0 = 28 \cdot 10^6 \text{ Hz}$

→ frequency on ground f = 28,000,730.17 Hz
→ difference of 730.17 Hz x 2 = 1460.34 Hz





Implementing a CW and FT4 beacon aboard this small satellite



Testing:

Doppler shift LEO satellite:

$$f = f_0 \left(1 + \frac{v}{c} \right)$$

v = speed of the approaching satellite 7823 m/s $c = velocity of the speed of light 3 . <math> 10^8 m/s$ $f_0 = 28 . 10^6 Hz$

→ frequency on ground f = 28,000,730.17 Hz

→ difference of 730.17 Hz x 2 = 1460.34 Hz

→ divided by 10 minutes pass = 2.435 Hz/s





Implementing a CW and FT4 beacon aboard this small satellite



Testing:

Doppler shift LEO satellite:

$$f = f_0 \left(1 + \frac{v}{c} \right)$$

v = speed of the approaching satellite 7823 m/s $c = velocity of the speed of light 3 . <math> 10^8 m/s$ $f_0 = 28 . 10^6 Hz$



$$\rightarrow$$
 divided by 10 minutes pass = 2.435 Hz/s

$$\rightarrow$$
 FT4 TX = 0.048s => 20.83 Hz





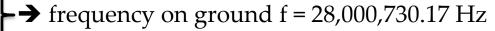
Implementing a CW and FT4 beacon aboard this small satellite



Testing:

Doppler shift LEO satellite:

$$f = f_0 \left(1 + \frac{v}{c} \right)$$



$$\rightarrow$$
 divided by 10 minutes pass = 2.435 Hz/s

$$\rightarrow$$
 FT4 TX = 0.048s => 20.83 Hz

 \rightarrow shift of 2.435 Hz/s / 20.83 Hz = 117 mHz



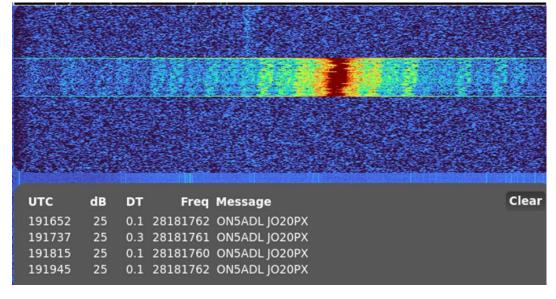


Implementing a CW and FT4 beacon aboard this small satellite



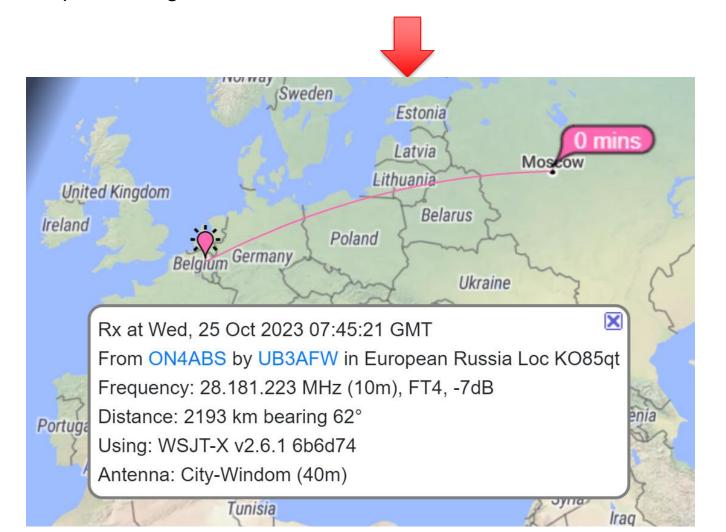
Testing:

```
Used payload='ON5ADL J020PX'
FT4 sequence started. Doppler delta=120 mHz
Enter payload (max 13 chars) >>200
Delta changed to 200 mHz !!
Enter payload (max 13 chars) >>on5adl jo20px
Used payload='ON5ADL J020PX'
FT4 sequence started. Doppler delta=200 mHz
Enter payload (max 13 chars) >>250
Delta changed to 250 mHz !!
Enter payload (max 13 chars) >>on5adl jo20px
Used payload='ON5ADL J020PX'
FT4 sequence started. Doppler delta=250 mHz
Enter payload (max 13 chars) >>
Used payload='ON5ADL J020PX'
FT4 sequence started. Doppler delta=250 mHz
Enter payload (max 13 chars) >>240
Delta changed to 240 mHz !!
Enter payload (max 13 chars) >>on5adl jo20px
Used payload='ON5ADL J020PX'
FT4 sequence started. Doppler delta=240 mHz
Enter payload (max 13 chars) >>
```



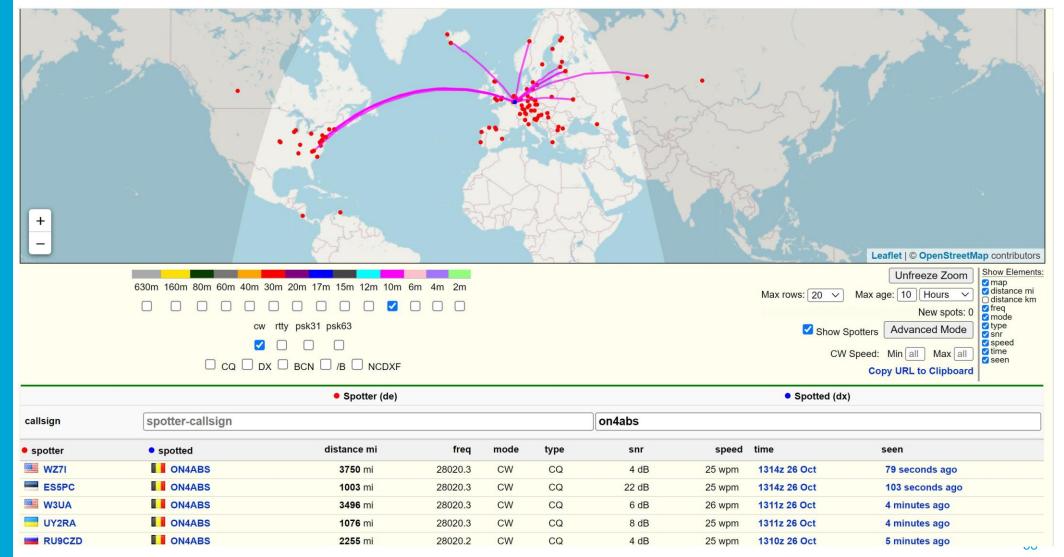
fUDelft





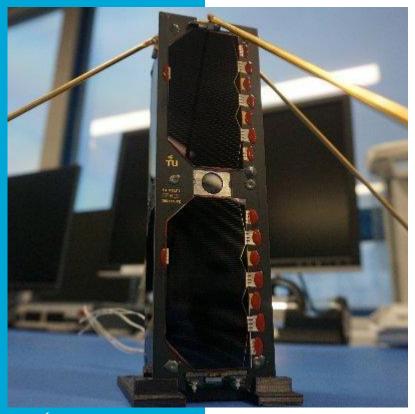


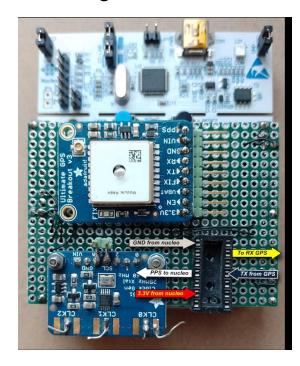




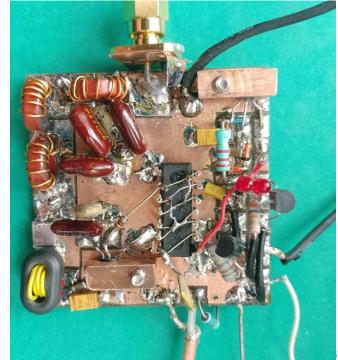
















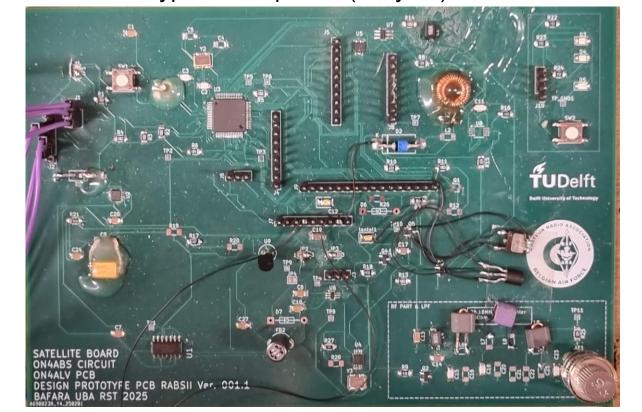
Implementing a CW and FT4 beacon aboard this small satellite



Prototype development (4 layers)







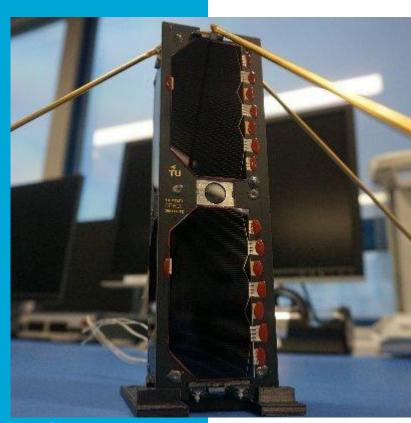






- Design a SMD protoype
- Functional testing
- Performance testing





















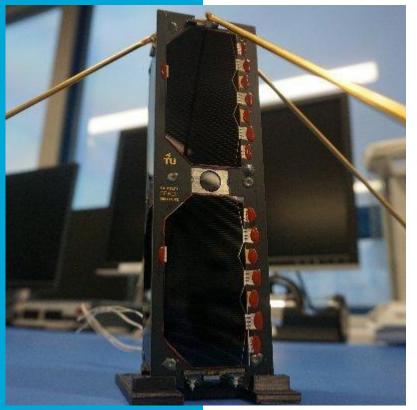


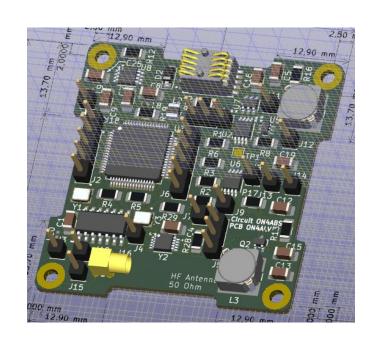


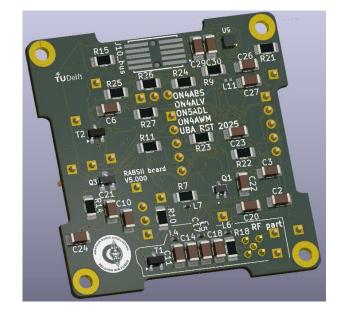
Implementing a CW and FT4 beacon aboard this small satellite



Multiple versions of 4,8 x 4,8 cm PCB (multilayer)











Implementing a CW and FT4 beacon aboard this small satellite



Multiple versions of 4,8 x 4,8 cm PCB (multilayer)















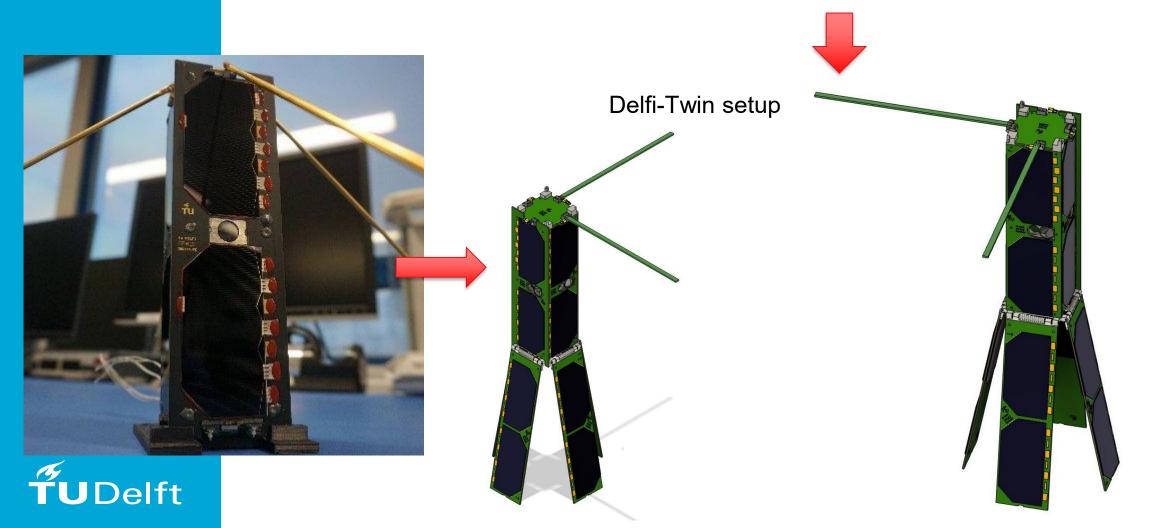
- Life cycle
- Vacuum testing
- Thermal cycling
- Radiation testing



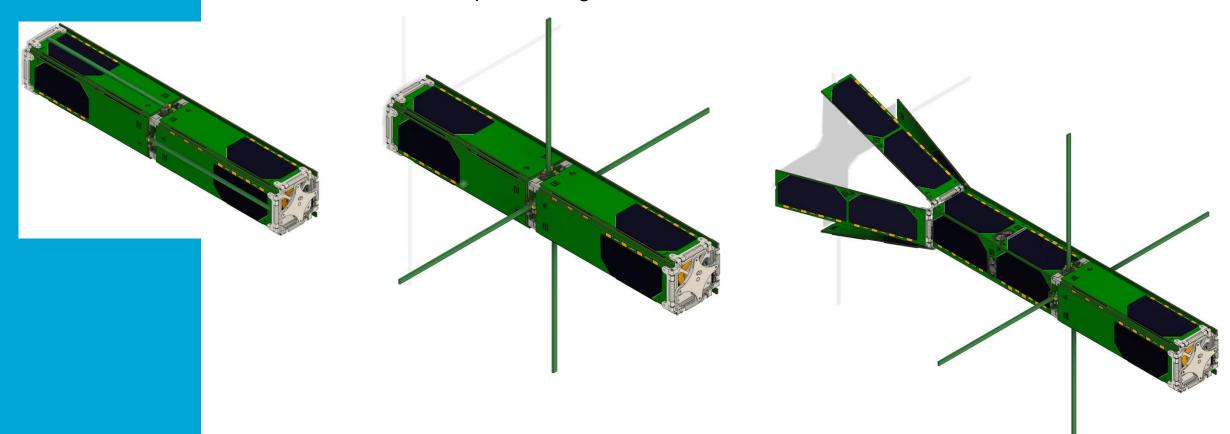






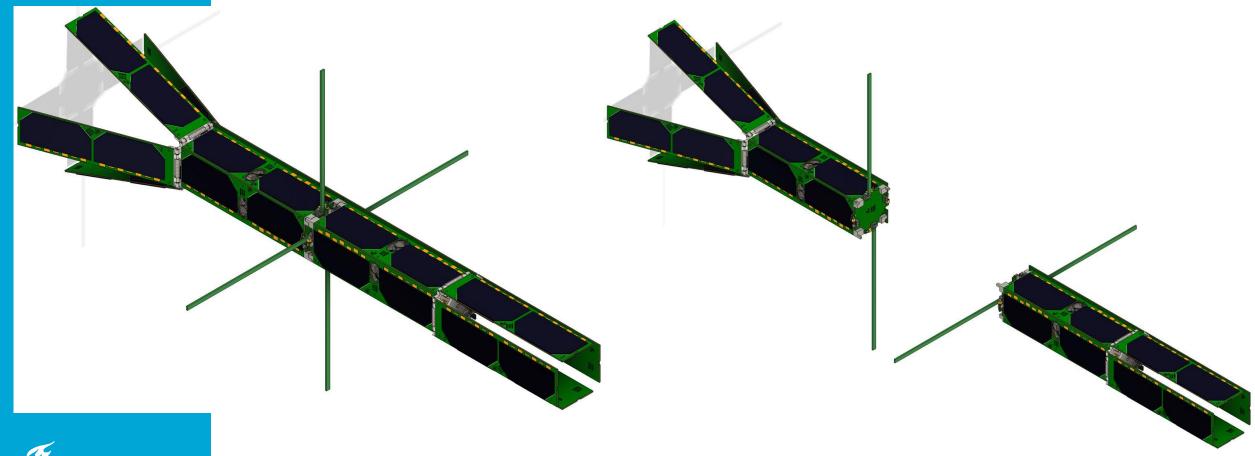








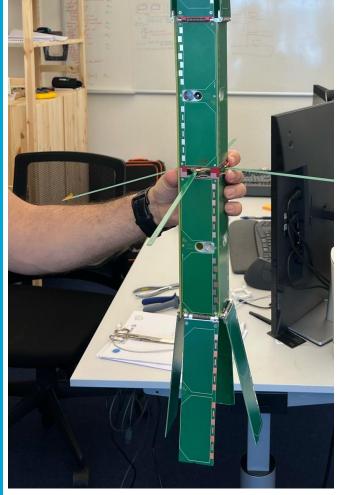


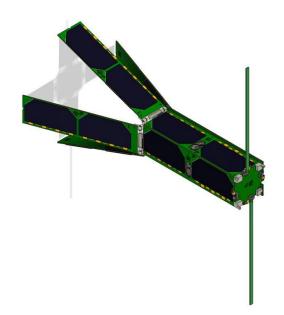


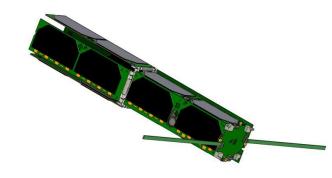






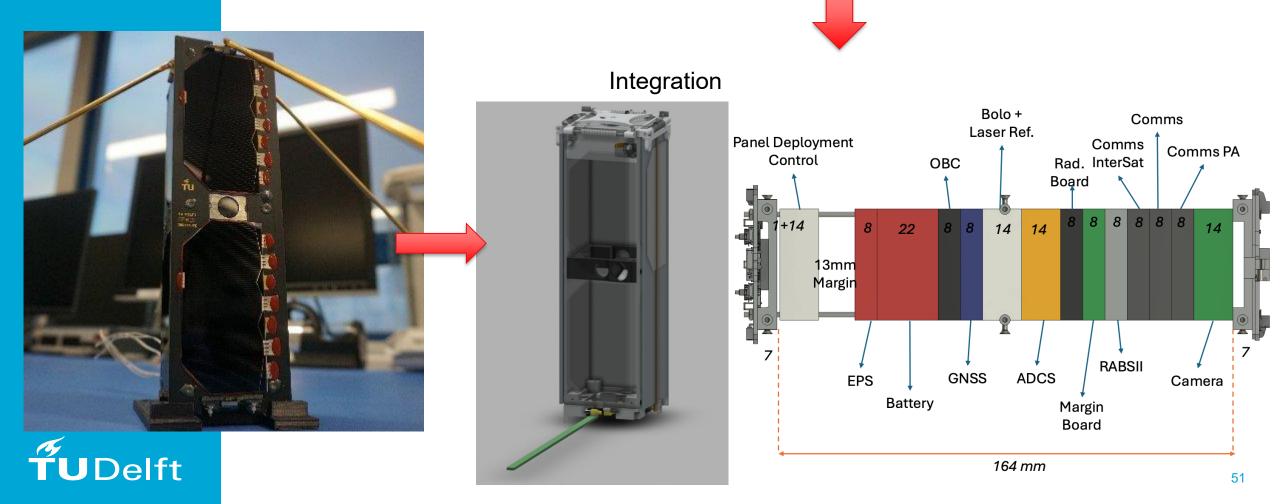




















Integrating the antenna aboard this small satellite



Challenges:

- Antenna design
- Deployment system

10m band: dipole → 5 m long → each leg: 2.5 m long

6m band: dipole → 3 m long → each leg: 1.5 m long

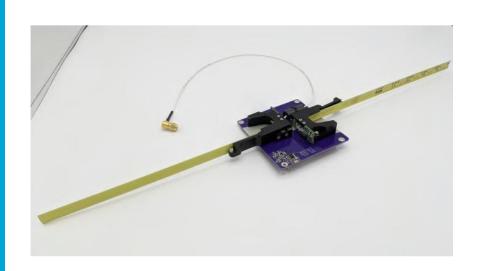


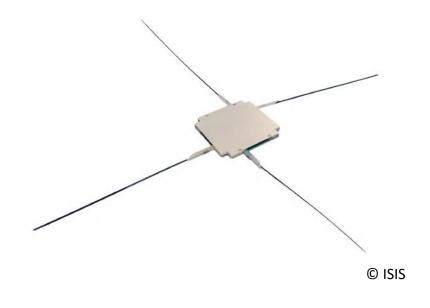


Integrating the antenna aboard this small satellite

10m band: dipole → 5 m long → each leg: 2.5 m long

6m band: dipole → 3 m long → each leg: 1.5 m long





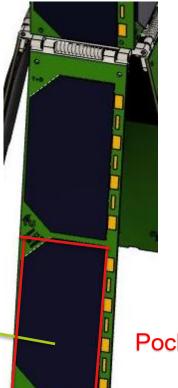




Integrating the antenna aboard this small satellite

10m band: dipole → 5 m long → each leg: 2.5 m long

6m band: dipole → 3 m long → each leg: 1.5 m long



Size: 2 x 2.5 m (28 MHz) 2 x 1.5 m (50 MHz)

Pocket size available: 8 cm x 4 cm x 2 mm





Integrating the antenna aboard this small satellite

10m band: dipole → 5 m long → each leg: 2.5 m long

6m band: dipole → 3 m long → each leg: 1.5 m long



Size: 2 x 2.5 m (28 MHz) 2 x 1.5 m (50 MHz)





Integrating the antenna aboard this small satellite



Nitinol: shape-memory alloy (SMA)





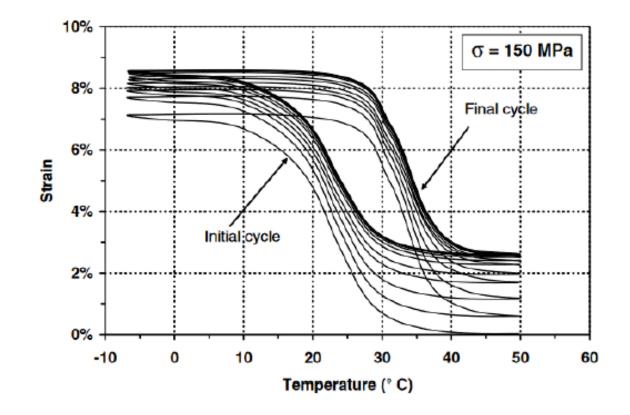




Integrating the antenna aboard this small satellite



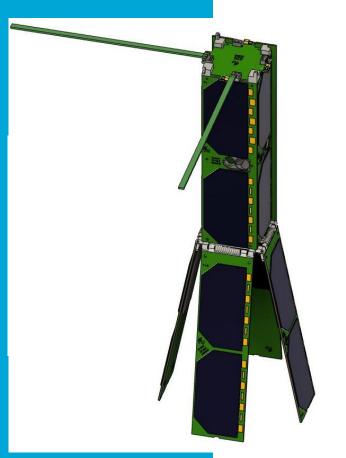
Nitinol: shape-memory alloy (SMA)







Integrating the antenna aboard this small satellite

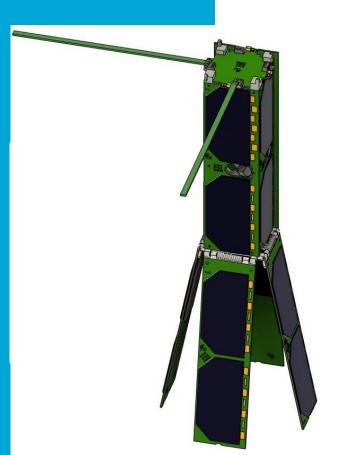


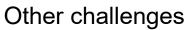
Nitinol: shape-memory alloy (SMA)







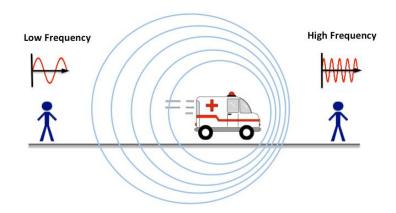


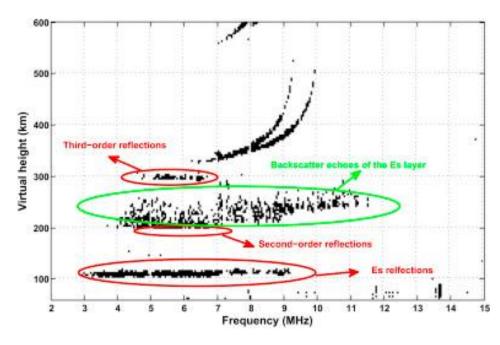




- Antenna design
- Deployment system
- Signal behavior
- Data analyses

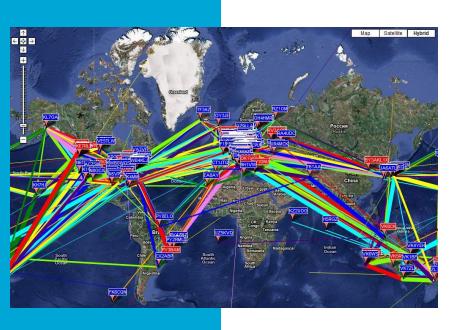












Other challenges



Challenges:

Data analyses

Signal reception → database



Receiving principle:

- Comparable with WSPR reception:
 - * use of Reversed Beacon Network (CW)
 - * use of WSJT-X OpenWebRX







Project timeline 2023

- Mission definition
 - Define payload mission requirements
 - Define satellite architecture
 - Payload listing
- Bus design iteration
 - Take lesson learned from Delfi-PQ
 - Improve bus design (SW, HW and operations)





Project timeline 2024-2027

- Mission definition
 - Select most promising payload options (2024)
 - Iterate on payload design (2025)
 - Flight model to be built (2025-2026)
- Bus design iteration
 - Design final bus concept





Project timeline 2024-2027

- 2025-2026
 - Satellite integration
 - Environmental tests
- 2026
 - Delivery for launch → TBC
- 2027
 - Launch → TBC





Further reading





Article

Concept of Sporadic E Monitoring Using Space-Based Low Power Multiple Beacon-Systems

Jurgen Vanhamel 1,2,*, Marc Berwaerts 3, Stefano Speretta 1,0 and Sevket Uludag 1,0

- ¹ TU Delft-Faculty of Aerospace Engineering, Kluyverweg 1, 2629 HS Delft, The Netherlands; s.speretta@tudelft.nl (S.S.); m.s.uludag@tudelft.nl (S.U.)
- Electronic Circuits and Systems, KU Leuven, Kleinhoefstraat 4, 2440 Geel, Belgium
- Union of Belgian Radio-Amateurs, RST-VZW, 3800 Sint-Truiden, Belgium; on4abs@myuba.be
- Correspondence: j.a.m.vanhamel@tudelft.nl

Abstract: Current monitoring systems to detect sporadic E use ground-based setups, ionosondes, and the network of GNSS satellites in order to assess the phenomenon of sporadic E. This paper aims to monitor sporadic E using a miniature space-based platform in an atypical way. The setup consists of multiple radio-amateur beacon systems aboard satellites, each having a specific modulation and transmission scheme. This Radio Amateur Beacon System for the Investigation of the Ionosphere (RABSII) is coupled to a GNSS receiver, revealing the location of the platform. Multiple beacon data streams are sequentially sent from a satellite platform towards the Earth. By receiving and comparing the Signal-to-Noise ratios of these streams using a dedicated ground-based radio-amateur network of receiving stations, the presence of sporadic E can be determined, and a location-based model can be built. The advantage of this miniaturized, low-power, low-cost instrument is its ability to be put on any satellite platform in the future in order to map sporadic E.

Link for free reading: https://www.mdpi.com/2073-4433/15/11/1306





Questions?



